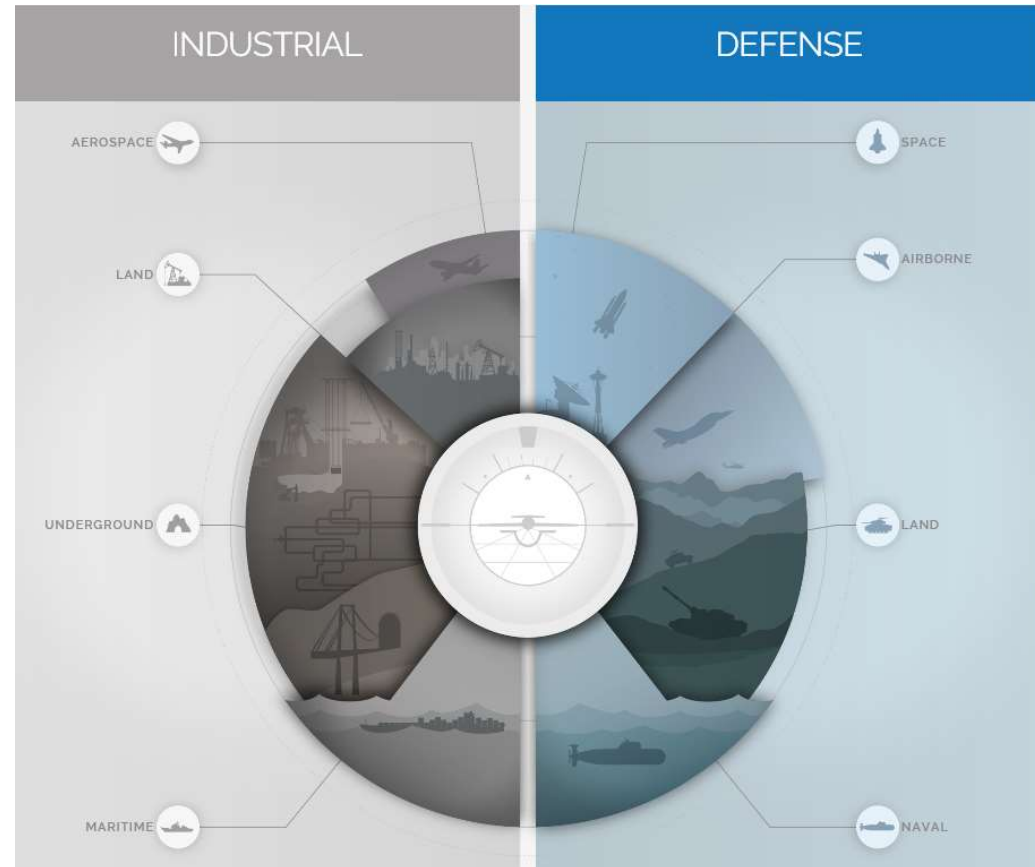




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Embedded Systems

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Director of Engineering

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High Performance Embedded GNSS INS (EGI) based on FOG Sensors Technology for Safety Critical Airborne Applications



Civitanavi Systems



- Vertically integrated Inertial Navigation Systems (INS) company, founded in 2012
- Design, development, certification and manufacturing of FOG (Fiber Optic Gyro) INS
- 100% engineering and production activities in-house

Headquarter in Pedaso (Fermo)



- Facilities in:
 - Pedaso (Fermo), Italy Headquarter (Adriatic east coast) – Systems & Sensors Design, Manufacturing
 - Ardea (Rome), Italy - Software Engineering, HW/SW Certification
- Civitanavi with a 40+ workforce of skilled engineers and experienced manufacturing team has delivered more than 350 since start-up

Subsidiary Office in Ardea (Rome)



Sensor Assembly Area



Sensor Calibration Equipment



Company Vision



“To become a worldwide supplier of Inertial Navigation, Geo-reference and Stabilization solutions provider, for commercial and dual use applications”



dual use *technology*

λβοιουπατ εσν λερ



Inertial Navigation System



- An Inertial Navigation System (INS) uses a self-contained navigation technique, based on the laws of classical mechanics, to estimate and track the orientation, position and velocity of the vehicle in which it is installed, with respect to a pre-defined reference system.
- The performance of an INS depends on many factors such as:
 - The technology for the realization of the sensors
 - The calibration process
 - The complexity of the mathematical model used for describing the process
 - The dependency of the sensor output from environmental conditions
 - Numerical integration of inertial data used to obtain the Navigation Data
- The development processes of devices for airborne applications must be constrained by the requirement of avionics standards such as RTCA standards DO-178 (software), DO-160 (environmental), DO-254 (complex hardware).

High Performance Embedded GNSS INS



- The Embedded GNSS INS (EGI) provides:
 - Euler Angles (True and Magnetic Heading, Pitch, Roll)
 - Aircraft Body Rates and Linear Accelerations
 - Inertial Position (Latitude, Longitude, Altitude) and Linear Velocities (VN, VE, VD)
 - Built-In Test diagnostics
- The EGI provides three simultaneous Navigation Solutions:
 - **Hybrid**: inertial measurements (Gyroscopes and Accelerometers) aided by the internal GPS receivers, when available
 - **Free Inertial**: inertial measurements (Gyroscopes and Accelerometers), without position or velocity aiding such as GPS.
 - **GPS-only**: It is an echo, in accordance with the Interface of the GPS data received



Sensor Calibration



The calibration procedure is necessary for various reasons:



- The sensor sensitivity axes usually do not coincide exactly with the body frame axes: this is due to manufacturing misalignments in the sensor mechanical assembly as well as imperfections of the sensor itself.
- Uncompensated Bias and Scale Factors: as their actual values usually differ from the nominal ones, they have to be determined.
- A fine calibration process requires a mechanical platform to precisely maneuver the INS.
- The developed INS calibration algorithm for compensating the systematic error sources is based on a two-steps approach:
 - thermal compensation
 - scale factor and mechanical misalignments estimation

INS Math Model and Kalman Filter



Standard INS Mathematical Model

$$\dot{\phi} = \frac{V_N}{M + h}$$

$$\dot{\lambda} = \frac{V_E}{(N + h)\cos(\phi)}$$

$$\dot{h} = -V_D$$

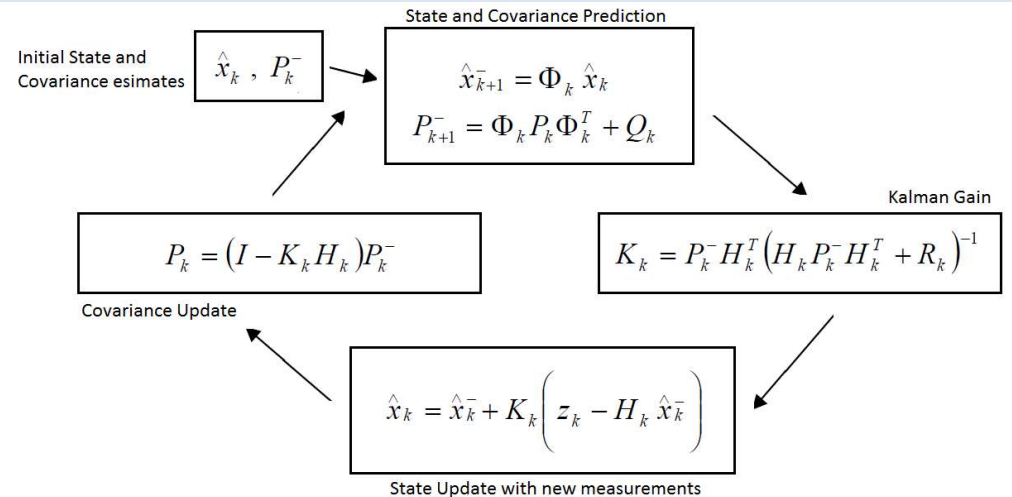
$\phi, \lambda = \text{lat, long}$
 $V = [V_N \ V_E \ V_D]$ velocities in NED frame
 $M, N = \text{meridian, transverse radius of Earth}$
 $h = \text{height}$

$$\dot{V} = C_b^n f_b (2\omega_{ie} + \omega_{en}) \times V + g_l^n$$

$$\dot{q} = \frac{1}{2} q (\omega_{ib} - C_n^b (\omega_{ie} + \omega_{en}))$$

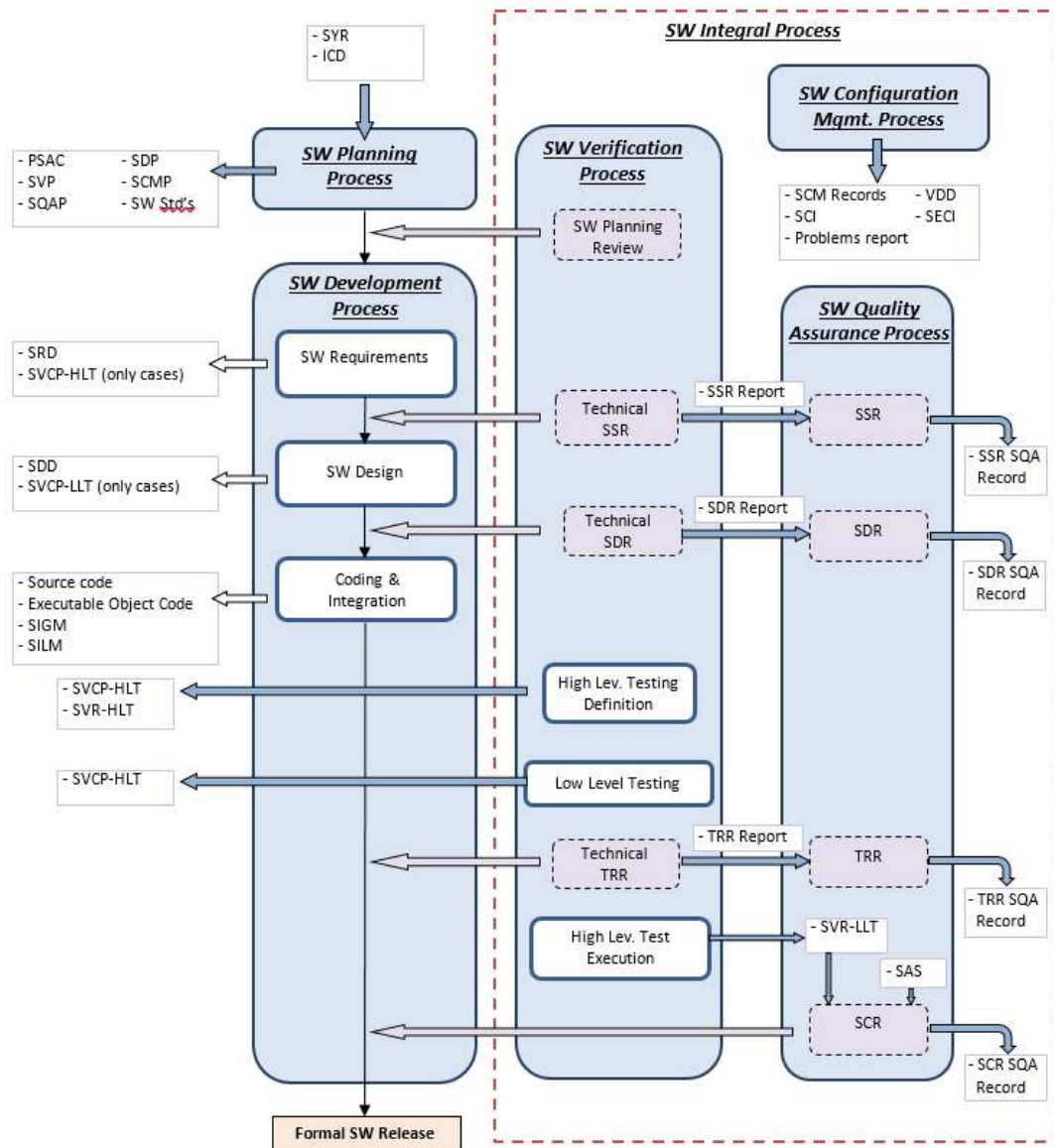
$\omega_{ie} = \text{Earth rate in NED frame}$
 $\omega_{en} = \text{transport rate (turn rate of NED wrt Earth-fixed frame)}$
 $\omega_{ib} = \text{true angular rate in body frame}$
 $g_l^n = \text{local gravity vector}$
 $f_b = \text{true acceleration in body frame}$
 $C_b^n, C_n^b = \text{transformation matrix body-to-NED and NED-to-body frame}$
 $q = \text{quaternion representation}$

Kalman Filter Scheme



- x_k is the state vector at time iteration k
- P_k is the state covariance matrix
- Q_k is the process noise matrix, estimated using the Inertial Sensors Error Budgets gathered from the Allan Variance analysis
- R_k is the measurement noise matrix, estimated exploiting the GPS Figure of Merit
- K_k is the Kalman gain
- H_k is the measurement model matrix
- Φ_k is the transition matrix (INS Math Model)

DO-178 SW Certification



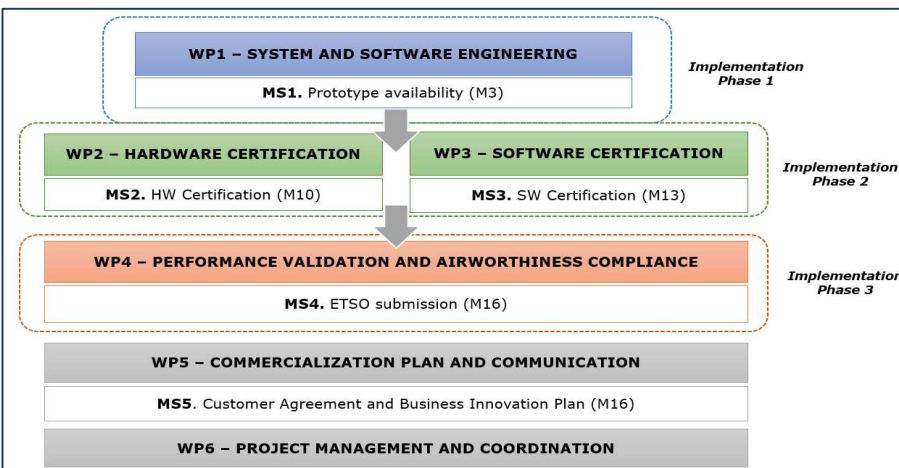
For all the software running on the Processing Unit in the INS, Civitanavi follows the design, development and verification rules of the RTCA/DO-178 standard:

- Planning Process
- Requirements Capture Process
- Conceptual Design Process
- Detailed Design
- Implementation Process
- Validation and Verification Process
- Conformity Review

H2020 EC Funded Project: NICENAV



Navigation-grade ITAR-free certifiable equipment for the navigation of air vehicle, based on FOG technology



La Civitanavi Systems di Pedaso sviluppa un equipaggiamento di navigazione inerziale con l'Unione Europea

23/12/2016 - L'azienda premiata come start up dell'anno. Civitanavi Systems ha sottoscritto ufficialmente con l'Unione Europea un importante contratto nell'ambito dell'iniziativa Horizon 2020 (SME Small and Medium Enterprise Innovation), il Programma Quadro per la Ricerca e l'Innovazione riservata alle Piccole Medie Imprese, per lo sviluppo di un innovativo apparato di Navigazione Inerziale per applicazioni aeronautiche.

Il progetto, denominato NICENAV, prevede lo sviluppo di un sofisticato equipaggiamento di navigazione inerziale basato su tecnologia FOG (Fiber Optic Gyroscope), di tipo non ITAR (International Traffic in Arms Regulations, Regolamenti in merito al commercio internazionale di armi) e certificato in ambito aeronautico (ETSO - European Technical Standard Order), che sposa l'obiettivo della Comunità Europea di promuovere una soluzione completamente europea, concorrenziale e non soggetta a vincoli di esportazione, in un mercato vasto e strategico quale quello della Navigazione Inerziale Aerospaziale, per applicazione su velivoli manned e unmanned.

L'assegnazione è avvenuta a valle di un processo di valutazione altamente competitivo condotto da un gruppo internazionale di esperti indipendenti che hanno individuato e premiato la soluzione tecnologica di maggiore impatto innovativo, riconoscendo anche in ambito internazionale, la capacità dell'azienda marchigiana di fare della ricerca avanzata e dell'innovazione tecnologica i principi fondamentali del proprio percorso di crescita e di sviluppo. **Solamente 7 aziende Italiane - tra cui Civitanavi Systems - su un totale di 528 provenienti da 21 Nazioni (pari allo 1.32%) sono state selezionate per la contrattualizzazione del progetto.**

"Per noi - ha affermato Andrea Pizzarulli, Amministratore unico di Civitanavi Systems - la partecipazione in Horizon 2020 rappresenta un grande successo, che ci consente di proporci e confermarci anche in ambiente internazionale come provider di soluzioni altamente innovative e di prodotti tecnologicamente competitivi nell'ambito Aerospaziale e anche in altre aree di mercato dedicate a specifici e differenti settori quali il Navale, il Terrestre, il Minerario e l'Oil&Gas".

Conclusions



- Civitanavi Embedded GNSS INS (EGI) for safety critical application has been presented
 - High performance
 - Modern FOG based technology
 - Advanced Navigation algorithm
 - Certified according to DO-178, DO-254 standards
 - Qualified according to DO-160 environmental/EMC tests
- Flight test activities have demonstrated the capability of the EGI

EGI System



DO-178, DO-254, DO-160 Qualification and Certification processes

